

ROLLADEN-SCHNEIDER Flugzeugbau GmbH LBA-Nr. EB - 4	Technische Mitteilung Nr. 6015	LS6 LS6-a	Blatt 1
			25.Feb.1988

- Gegenstand : Ergänzung des Flughandbuchs und Änderung des Verfahrens zur Bestimmung des Höchstzulässigen Gewichts der Nichttragenden Teile. (Änderung des Wartungshandbuchs)
- Betroffen : Segelflugzeugmuster LS6, Baureihen LS6 und LS6-a, alle Werknummern
- Dringlichkeit : wahlweise
- Vorgang : Das Verfahren zur Bestimmung des höchstzulässigen Gewichts der Nichttragenden Teile wurde geändert (analog LS6-b):
Das höchstzulässige Gewicht der Nichttragenden Teile wird in Abhängigkeit von Leergewicht und Leergewichts-Schwerpunktlage aus einer Tabelle entnommen.
- Maßnahmen und Material : LS6: Austausch von Blatt 0-5 und 1-5 des Flughandbuchs gegen Ausgabe 25.Feb.88 (Revision 3)
Austausch der Blätter 0-3 und 2-1 bis 2-5 und 14-4 des Wartungshandbuchs gegen Ausgabe 25.Feb.88 (Revision 4)
LS6-a: Austausch von Blatt 0-5 und 1-5 des Flughandbuchs gegen Ausgabe 25.Feb.88 (Revision 2)
Austausch der Blätter 0-3 und 2-1 bis 2-5 und 14-4 des Wartungshandbuchs gegen Ausgabe 25.Feb.88 (Revision 2)
- Gewicht und S.P.-Lagen : nicht betroffen
- Hinweise : Bei rückwärtiger Leergewicht-Schwerpunktlage und hohem Leergewicht empfiehlt sich eine Kontrolle der Höchstzuladung für das Cockpit. War diese durch das Höchstgewicht der Nichttragenden Teile begrenzt, dann wird sie nach dem neuen Verfahren höher.

Durchführung durch den Halter.
Bescheinigung der Durchführung im Rahmen der nächsten Jahresnachprüfung durch einen Prüfer Klasse 3 im Bordbuch sowie im TM-LTA-Durchführungsbeleg, Wartungshandbuch Blatt 14-1.
Kontrolle der Höchstzuladung im Cockpit und Eintrag des neuen Werts im Flughandbuch Blatt 1-6 durch einen Prüfer Klasse 3.

LBA-anerkannt :



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25. FEB. 1988

Erstellt: 25. FEB. 1988

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Geprüft: 25. FEB. 1988

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	No. 6015	LS6-a	Edition: 25.Feb.88

SUBJECT : Modification of the calculation procedure to establish maximum permissible weight of non-lifting parts (Modification of Instructions for Continued Airworthiness).

EFFECTIVITY : Sailplane model LS6, versions LS6 and LS6-a, all serial numbers

ACCOMPLISHMENT : Optional

REASON : Calculation procedure to establish maximum permissible weight of non-lifting parts has been modified analogous to the LS6-b procedure: Maximum weight of non-lifting parts is taken from table in relation to empty weight and empty weight C.G. position.

MATERIAL and INSTRUCTIONS : Exchange following pages:
LS6: Flight Manual:
(Gen./USA Ed.) pages 1-1, 1-2, 1-4, 2-4, 6-1 to 6-7 Edition 25.Feb.1988 (Revision 4)
(USA Edition) pages 1-1, 1-2, 1-4, 2-4, 6-1 to 6-7 Edition 25.Feb.1988 (Revision 3)
(Italy Edition) pages 1-1, 1-2, 1-4, 2-4, 6-1 to 6-7 Edition 25.Feb.1988 (Revision 1)
LS6: Instructions for Continued Airworthiness:
(Gen./USA Ed.) pages 1-1, 1-2, 8-4 Edition 25.Feb.1988 (Revision 1)
(Italy Edition) pages 1-1, 1-2, 8-4 Edition 25.Feb.1988 (Revision 1)
LS6-a: Flight Manual:
(Gen./USA Ed.) pages 1-1, 1-2, 1-4, 2-4, 6-1 to 6-7 Edition 25.Feb.1988 (Revision 3)
(Italy Edition) pages 1-1, 1-2, 1-4, 2-4, 6-1 to 6-7 Edition 25.Feb.1988 (Revision 1)
LS6-a: Instructions for Continued Airworthiness:
(Gen./USA Ed.) pages 1-1, 1-2, 8-4 Edition 25.Feb.1988 (Revision 3)
(Italy Edition) pages 1-1, 1-2, 8-4 Edition 25.Feb.1988 (Revision 1)

WEIGHT AND BALANCE : Not applicable

REMARKS : For rearward C.G. positions and high empty weight the maximum cockpit load should be checked. If this was reduced by the maximum weight of non-lifting parts, with this new method the cockpit load becomes higher.

Exchange of pages by operator.
Certification of accomplishment with next annual inspection by licensed inspector in logbook and on TB-AD-Accomplishment List, page 8-1 of Instructions for Continued Airworthiness (Maintenance Manual).
Checking of maximum cockpit load and entry of new value into Flight Manual (page 9-1) by licensed inspector.

LBA-approved :



Erstellt: 25. FEB. 1988


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1.1 LOG OF REVISIONS

Revision No.	Pages affected	Description	LBA-approval signature	Date
1	1-1, 1-2, 4-2, 4-12, 4-13, 7-1	Hydraulic damper for aileron system (TB 6010/11, LBA-AD 86-140/2)		
2	1-1, 1-2, 2-5, 3-1, 3-2, 4-4, 4-7 to 4-10 9-3	Updating of Manual (TB 6014)		
3	1-1, 1-2, 1-4, 2-4, 6-1 to 6-7	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)	 <i>[Signature]</i>	8. Jan 1988

LS6 Manuals can be ordered from:
 Rolladen-Schneider Flugzeugbau GmbH
 Mühlstrasse 10
 D-6073 Egelsbach

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1.1 LOG OF REVISIONS

Revision No.	Pages affected	Description	LBA-approval signature	Date
1	1-1, 1-2, 1-6, 2-4, 2-5, 4-7, 5-1, 6-2, 6-5, 6-7, 7-2, 9-2, 9-5	Amount of water ballast and forward empty weight C.G. limit table according to TB 6007		
2	1-1, 1-2, 4-2, 4-12, 4-13, 7-1	Hydraulic damper for aileron system (TB 6010/11, LBA-AD 86-140/2)		
3				
4	1-1, 1-2, 1-4, 2-4, 6-1 to 6-7	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)		25. Feb. 1988

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2.4 MASS (WEIGHT)

Maximum take-off mass with water ballast 525 kg (1157 lbs)
without water ballast 380 kg (838 lbs)

Maximum landing mass 525 kg (1157 lbs)

Maximum mass of all non-lifting parts 230 to 245 kg (507-540 lbs)

Value must be determined according to table page 6-4, related to empty weight and empty weight C.G. position.

The term "non-lifting parts" includes the following: fuselage inclusive permanently fitted equipment, canopy, battery in baggage compartment and/or tail fin and main pins plus maximum cockpit load.

Maximum water ballast mass 140 kg (309 lbs)
Loading instructions see page 9-2.

Maximum mass in Baggage Compartment 5 kg (11 lbs)
Loading instructions see page 4-6.

Maximum Cockpit Load 110 kg (242 lbs)

The term "Cockpit Load" includes the following:
Pilot, parachute, baggage and temporary equipment.

Maximum cockpit load may be limited by mass of non-lifting parts. See entry on page 9-1.

Minimum Cockpit Load for club use (recommended)

Pilot and parachute 70 kg (154 lbs)
No baggage, no temporary equipment, no trim ballast

Pilot and parachute 55 kg (121 lbs)
3 trim weights fitted, no baggage, no temporary equipment

One trim weight (2.5 kg, 6 lbs) corresponds to 5 kg (11 lbs) of pilot mass.

If the sailplane does not fly in a club, it may be trimmed for higher minimum cockpit load. See instructions on page 9-4.

For minimum cockpit load see entry on page 9-1.

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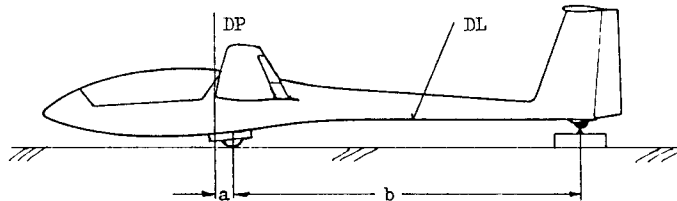
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6.1 WEIGHING PROCEDURE

To determine inflight C.G. position, the empty weight C.G. position must be known.

1. Determine total weight by weighing all parts and adding together. For inflight C.G. position, the pilot's weight must be added too.
2. Assemble sailplane. For inflight C.G., the pilot must be seated in the sailplane.
3. Raise tail on weighing machine until datum line is level using wooden blocks or adjustable jack. (Check with leveling gauge)
4. Determine gross tail weight.
5. Measure distance from tail support to center of landing gear axis.
6. Using plumb lead, determine points on floor perpendicular to left and right datum points, and points on floor perpendicular to center of landing gear axis. Measure distance <a> from wheel axis to datum point.



DATUM LINE <DL>: Under side of fuselage boom placed horizontal
 DATUM POINT <DP>: Leading edge of wing at root

7. Determine tare tail weight (Weight of auxiliary support used under 3)
8. Calculate nett tail weight = gross tail weight minus tare tail weight (the pilot was included):
9. Calculate empty weight C.G. position:

$$X_{cg} = \frac{\text{nett tail weight} * b}{\text{empty weight}} + a$$

10. When a battery is fitted in the vertical tail fin, weighing must be done in this configuration. Weigh tail fin battery separately. (Maximum 2.6 kg <5.7 lbs>).
11. Calculate loading limits according to page 6-2.

Form for Weighing Report see Instructions for Continued Airworthiness, page 8-4

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6.2 CALCULATION OF LOADING LIMITS

1. Determine Minimum Cockpit Load from Table "Empty Weight C.G. Position", page 6-6 or 6-7.
When being used in a club, Minimum Cockpit Load should be 70 kg (154 lbs). If it is higher, permanent ballast may be fitted under the forward seat portion, see page 9-1.

Finally resulting Minimum Cockpit Load should be entered in the following places:

- 1) in weighing report of inspection
 - 2) in Flight Manual page 9-1
 - 3) under instrument panel cover
 - 4) on Data Placard in cockpit
2. Maximum approved Weight of Non-lifting Parts may vary between 230 kg and 245 kg (507 lbs to 540 lbs).
In contrast to methods used up to now, Maximum Weight of Non-lifting Parts can be determined in relation to Empty Weight and Empty Weight C.G. Position according to table on page 6-4 and 6-5. See also example on page 6-3.

Maximum Weight of Non-lifting Parts should be entered into Weighing Report.

3. Determine Maximum approved Cockpit Load from table "Empty Weight C.G. Position", page 6-6 and 6-7.
Maximum Cockpit Load normally is 110 kg (242 lbs), as given in empty weight C.G. table. It may be lower due to trim conditions, excessive equipment or repairs.
Calculate Maximum Cockpit Load on weighing report, see also example on page 6-3.

Resulting Maximum Cockpit Load should be entered in the following places:

- 1) in weighing report of inspection
 - 2) in Flight Manual, page 9-1
 - 3) on Data Placard in cockpit
4. Empty weight (perhaps increased by weight of permanently fitted trim ballast) should be entered in the following places:
 - 1) in weighing report of inspection
 - 2) in Flight Manual, page 9-1 for calculation of maximum permissible water ballast weight

For permanent installation of trim ballast weights see page 9-5.

Form for Weighing Report see Instructions for Continued Airworthiness, page 8-4

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6.2 CALCULATION OF LOADING LIMITS continued

EXAMPLE for calculation of loading limits:

See also Instructions for Continued Airworthiness, form Weighing Report

1) Minimum Cockpit Load

For empty weight of 255 kg <562 lbs> and empty weight C.G. position 665 mm <26.18 in>, the Minimum Cockpit Load according to table page 6-6 and 6-7 is 75 kg <165 lbs> (Limit value ...667 mm <26.26 in> greater than actual value 665 mm <26.181 in>)

2) Maximum Weight of Non-lifting Parts

Maximum weight of non-lifting parts at empty weight of 255 kg <562 lbs> and empty weight C.G. position of 665 mm <26.181 in> is according to table page 6-4 and 6-5
..... 245 kg <540 lbs>

3) Maximum permissible Cockpit Load

FUSELAGE with complete equipment,
battery, canopy and main pins 120.6 kg <266 lbs>
HORIZONTAL TAIL UNIT 6.3 kg < 14 lbs>
COCKPIT LOAD (Maximum 110 kg <242 lbs> 110 kg <242 lbs>
Weight of Non-lifting Parts 236.9 kg <522 lbs>

MAXIMUM COCKPIT LOAD (Max. 110 kg <242 lbs>) 110 kg <242 lbs>

Maximum all-up weight 525 kg <1157 lbs>

Erstellt: 25. FEB. 1988 <i>Lauke</i>	Geprüft: 25. FEB. 1988 <i>Whapla</i>
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6.3 CALCULATION OF MAXIMUM WEIGHT OF NON-LIFTING PARTS

Maximum weight of non-lifting parts of 245 kg <540 lbs> must be reduced in relation to empty weight and empty weight C.G. position according to table below. (For lbs / inch values see following page)

Example: For empty weight C.G. position of 639 mm <25.16 in> and empty weight of 249 kg <549 lbs> the permissible weight of non-lifting parts is 232 kg <533 lbs>.

Empty Weight G <kg>	Empty weight C.G. Position Xs <mm>											
	from 540 to 559	from 560 to 579	from 580 to 599	from 600 to 619	from 620 to 639	from 640 to 659	from 660 to 679	from 680 to 699	from 700 to 719	from 720 to 739	from 740 to 759	from 760 to 779
246	230	230	230	231	232	233	234	236	237	238	239	240
247	230	230	230	231	232	233	235	236	237	238	239	241
248	230	230	230	231	232	233	235	236	237	238	240	241
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258	230	230	231	232	233	234	236	237	238	239	241	242
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275	230	231	232	233	235	236	237	239	240	241	243	244
276	230	231	232	234	235	236	238	239	240	242	243	244
277	230	231	232	234	235	236	238	239	240	242	243	244
278	230	231	232	234	235	236	238	239	241	242	243	245
279	230	231	232	234	235	237	238	239	241	242	243	245
280	230	231	233	234	235	237	238	239	241	242	243	245

6.3 CALCULATION OF MAXIMUM WEIGHT OF NON-LIFTING PARTS continued

Maximum weight of non-lifting parts of 540 lbs <245 kg> must be reduced in relation to empty weight and empty weight C.G. position according to table below. (For kg / mm values see preceding page)

Example: For empty weight C.G. position of 25.16 in <639 mm> and empty weight of 549 lbs <249 kg> the permissible weight of non-lifting parts is 511 lbs <234 kg>.

Empty weight C.G. Position Xs <inches>

Empty Weight G<lbs>	from	from	from	from	from	from	from	from	from	from	from	from
	21.26 to 22.01	22.02 to 22.80	22.81 to 23.58	23.59 to 24.37	24.38 to 25.16	25.17 to 25.95	25.98 to 26.73	26.77 to 27.52	27.53 to 28.31	28.32 to 29.09	29.13 to 29.88	29.92 to 30.70
542	507	507	507	509	511	513	515	520	522	524	526	529
544	507	507	507	509	511	513	518	520	522	524	526	531
546	507	507	507	509	511	513	518	520	522	524	529	531
548	507	507	507	509	511	515	518	520	522	524	529	531
551	507	507	507	509	511	515	518	520	522	524	529	531
553	507	507	507	509	513	515	518	520	522	524	529	531
555	507	507	507	509	513	515	518	520	524	526	529	531
557	507	507	507	509	513	515	518	520	524	526	529	531
559	507	507	507	511	513	515	518	522	524	526	529	531
562	507	507	507	511	513	515	518	522	524	526	529	533
564	507	507	509	511	513	515	518	522	524	526	529	533
566	507	507	509	511	513	515	520	522	524	526	531	533
568	507	507	509	511	513	515	520	522	524	526	531	533
570	507	507	509	511	513	518	520	522	524	529	531	533
573	507	507	509	511	513	518	520	522	524	529	531	533
575	507	507	509	511	513	518	520	522	526	529	531	533
577	507	507	509	511	515	518	520	522	526	529	531	533
579	507	507	509	511	515	518	520	524	526	529	531	535
582	507	507	509	511	515	518	520	524	526	529	531	535
584	507	507	509	513	515	518	520	524	526	529	533	535
586	507	507	509	513	515	518	522	524	526	529	533	535
588	507	507	509	513	515	518	522	524	526	531	533	535
590	507	507	511	513	515	518	522	524	526	531	533	535
593	507	507	511	513	515	520	522	524	526	531	533	535
595	507	507	511	513	515	520	522	524	529	531	533	537
597	507	507	511	513	515	520	522	524	529	531	533	537
599	507	509	511	513	518	520	522	524	529	531	533	537
601	507	509	511	513	518	520	522	526	529	531	535	537
604	507	509	511	513	518	520	522	526	529	531	535	537
606	507	509	511	513	518	520	522	526	529	531	535	537
608	507	509	511	515	518	520	524	526	529	533	535	537
610	507	509	511	515	518	520	524	526	529	533	535	537
612	507	509	511	515	518	520	524	526	531	533	535	540
615	507	509	511	515	518	522	524	526	531	533	535	540
617	507	509	513	515	518	522	524	526	531	533	535	540

6.4 EMPTY WEIGHT C.G. TABLE <mm, kg>

For in, lbs values see following page

Empty Weight <kg>	Maximum Cockpit Load to Minimum Cockpit Load <kg>				
	110 - 70	110 - 75	110 - 80	110 - 85	110 - 90
246	659 - 659	659 - 677	659 - 695	659 - 713	659 - 730
247	657 - 658	657 - 676	657 - 694	657 - 711	657 - 729
248	656 - 657	656 - 675	656 - 693	656 - 710	656 - 728
249	654 - 656	654 - 674	654 - 691	654 - 709	654 - 726
250	653 - 655	653 - 673	653 - 690	653 - 708	653 - 725
251	651 - 654	651 - 672	651 - 689	651 - 706	651 - 724
252	649 - 653	649 - 671	649 - 688	649 - 705	649 - 723
253	648 - 652	648 - 669	648 - 687	648 - 704	648 - 721
254	646 - 651	646 - 668	646 - 686	646 - 703	646 - 720
255	645 - 650	645 - 667	645 - 685	645 - 702	645 - 719
256	643 - 649	643 - 666	643 - 683	643 - 700	643 - 717
257	642 - 648	642 - 665	642 - 682	642 - 699	642 - 716
258	640 - 647	640 - 664	640 - 681	640 - 698	640 - 715
259	639 - 646	639 - 663	639 - 680	639 - 697	639 - 714
260	637 - 645	637 - 662	637 - 679	637 - 696	637 - 713
261	636 - 644	636 - 661	636 - 678	636 - 695	636 - 711
262	634 - 643	634 - 660	634 - 677	634 - 694	634 - 710
263	633 - 642	633 - 659	633 - 676	633 - 692	633 - 709
264	631 - 642	631 - 658	631 - 675	631 - 691	631 - 708
265	630 - 641	630 - 657	630 - 674	630 - 690	630 - 707
266	628 - 640	628 - 656	628 - 673	628 - 689	628 - 706
267	627 - 639	627 - 655	627 - 672	627 - 688	627 - 704
268	625 - 638	625 - 654	625 - 671	625 - 687	625 - 703
269	624 - 637	624 - 653	624 - 670	624 - 686	624 - 702
270	623 - 636	623 - 653	623 - 669	623 - 685	623 - 701
271	621 - 635	621 - 652	621 - 668	621 - 684	621 - 700
272	620 - 634	620 - 651	620 - 667	620 - 682	620 - 699
273	619 - 634	619 - 650	619 - 666	619 - 682	619 - 698
274	617 - 633	617 - 649	617 - 665	617 - 681	617 - 697
275	616 - 632	616 - 648	616 - 664	616 - 680	616 - 696
276	614 - 631	614 - 647	614 - 663	614 - 679	614 - 694
277	613 - 630	613 - 646	613 - 662	613 - 678	613 - 693
278	611 - 629	611 - 645	611 - 661	611 - 677	611 - 692
279	610 - 629	610 - 644	610 - 660	610 - 676	610 - 691
280	608 - 628	608 - 643	608 - 659	608 - 675	608 - 690

6.4 EMPTY WEIGHT C.G. TABLE <in, lbs>

For mm, kg values see preceding page

Empty Weight <lbs>	Maximum Cockpit Load to Minimum Cockpit Load <lbs>				
	232 - 154	232 - 165	232 - 176	232 - 187	232 - 198
542	25.945- 25.945	25.945- 26.654	25.945- 27.362	25.945- 28.071	25.945- 28.740
544	25.866- 25.906	25.866- 26.614	25.866- 27.323	25.866- 27.992	25.866- 28.701
546	25.827- 25.866	25.827- 26.575	25.827- 27.283	25.827- 27.953	25.827- 28.661
548	25.748- 25.827	25.748- 26.535	25.748- 27.205	25.748- 27.913	25.748- 28.583
551	25.709- 25.787	25.709- 26.496	25.709- 27.165	25.709- 27.874	25.709- 28.543
553	25.630- 25.748	25.630- 25.400	25.630- 27.126	25.630- 27.795	25.630- 28.504
555	25.551- 25.709	25.551- 26.417	25.551- 27.087	25.551- 27.756	25.551- 28.465
557	25.512- 25.669	25.512- 26.339	25.512- 27.047	25.512- 27.717	25.512- 28.386
559	25.433- 25.630	25.433- 26.299	25.433- 27.008	25.433- 27.677	25.433- 28.346
562	25.394- 25.591	25.394- 26.260	25.394- 26.969	25.394- 27.638	25.394- 28.307
564	25.315- 25.551	25.315- 26.220	25.315- 26.890	25.315- 27.559	25.315- 28.228
566	25.276- 25.512	25.276- 26.181	25.276- 26.850	25.276- 27.520	25.276- 28.189
568	25.197- 25.472	25.197- 26.142	25.197- 26.811	25.197- 27.480	25.197- 28.150
570	25.157- 25.433	25.157- 26.102	25.157- 26.772	25.157- 27.441	25.157- 28.110
573	25.079- 25.394	25.079- 26.063	25.079- 26.732	25.079- 27.402	25.079- 28.071
575	25.039- 25.354	25.039- 26.023	25.039- 26.693	25.039- 27.362	25.039- 27.992
577	24.961- 25.315	24.961- 25.984	24.961- 26.654	24.961- 27.323	24.961- 27.953
579	24.921- 25.276	24.921- 25.945	24.921- 26.614	24.921- 27.244	24.921- 27.913
582	24.843- 25.276	24.843- 25.906	24.843- 26.575	24.843- 27.205	24.843- 27.874
584	24.803- 25.236	24.803- 25.866	24.803- 26.535	24.803- 27.165	24.803- 27.835
586	24.724- 25.197	24.724- 25.827	24.724- 26.496	24.724- 27.126	24.724- 27.795
588	24.685- 25.157	24.685- 25.787	24.685- 25.400	24.685- 27.087	24.685- 27.717
590	24.606- 25.118	24.606- 25.748	24.606- 26.417	24.606- 27.047	24.606- 27.677
593	24.567- 25.079	24.567- 25.709	24.567- 26.378	24.567- 27.008	24.567- 27.638
595	24.528- 25.039	24.528- 25.709	24.528- 26.339	24.528- 26.969	24.528- 27.598
597	24.449- 25.000	24.449- 25.669	24.449- 26.299	24.449- 26.929	24.449- 27.559
599	24.409- 24.961	24.409- 25.630	24.409- 26.260	24.409- 26.850	24.409- 27.520
601	24.370- 24.961	24.370- 25.591	24.370- 26.220	24.370- 26.850	24.370- 27.480
604	24.291- 24.921	24.291- 25.551	24.291- 26.181	24.291- 26.811	24.291- 27.441
606	24.252- 24.882	24.252- 25.512	24.252- 26.142	24.252- 26.772	24.252- 27.402
608	24.173- 24.843	24.173- 25.472	24.173- 26.102	24.173- 26.732	24.173- 27.323
610	24.134- 24.803	24.134- 25.433	24.134- 26.063	24.134- 26.693	24.134- 27.283
612	24.055- 24.764	24.055- 25.394	24.055- 26.023	24.055- 26.654	24.055- 27.234
615	24.016- 24.764	24.016- 25.354	24.016- 25.984	24.016- 26.614	24.016- 27.205
617	23.937- 24.723	23.937- 25.315	23.937- 25.945	23.937- 26.575	23.937- 27.165

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1.1 LOG OF REVISIONS

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1	1-1, 1-2, 2-1, 3-8, 8-8, 8-9, 8-11, 8-12	Hydraulic damper for aileron system (TB 6010/11, LBA-AD 86-140/2)		
2	1-1, 1-2, 8-4	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)		

LS6 Manuals can be ordered from:
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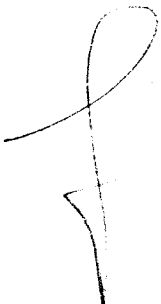
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1.1 LOG OF REVISIONS

Revision No.	Pages affected	Description	LBA-approval signature	Date
1	1-1, 1-2, 8-4	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)		
				

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Serial No. _____ Reg.Signs _____ Date: _____ **WEIGHING REPORT**

COMPONENT WEIGHTS (Check when equipment altered or every fourth year)

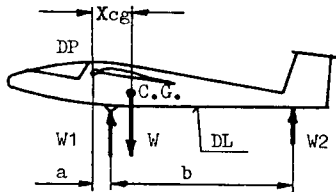
	Dimension		Dimension	
Right Wing		Maximum all-up Weight (1157 lbs)	525	kg
Left Wing		Maximum Weight of Non-lifting Parts at W= . . . kg/lbs and Xcg= . . . mm/in according to table on page 6-4/6-5 of Flight Manual.		
Fuselage + Equipment + Main Pins + Canopy		Fuselage (as on left side)		
Horizontal Tail		Horizontal tail		
Empty Weight <W>		Cockpit Load (max.110 kg or 242 lbs)		
BATTERY None		Weight of Non-lifting Parts		
POSITION Baggage Compartment		FIXED BALLAST POS. . . . kg/lbs Tail Fin/. . .		
Tail Fin				

WEIGHING AND EMPTY WEIGHT C.G. DETERMINATION (Check when equipment altered or every fourth year)

Technical data according to TCDS: Datum Point <DP>: Leading edge of wing at root
 Datum Line <DL>: Under side of fuselage boom horizontal

Empty Weight <W>		Distance Wheel axis - Datum Point <a>		
Nett Tail Weight <W2>		Distance Wheel axis - Tail support 		

NOTE: See also Flight Manual Chapter 6



$$\frac{W2 * b}{W} + a = Xcg$$

* + = kg / lbs

State dimensions used. Redetermine distances a and b, because of possibly altered suspension level.

Empty Weight C.G. Range according to Flight Manual, Chapter 6 from . . . mm/in to . . . mm/in at

Empty Weight <W> . . . kg/lbs yields permissible Cockpit Load Range

from . . . kg/lbs to . . . kg/lbs. Weighed and calculated C.G. position is within permissible limits.

MINIMUM COCKPIT LOAD: _____ kg/lbs **MAXIMUM COCKPIT LOAD:** _____ kg/lbs


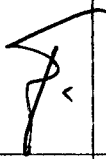
Weight and Balance Placard and Minimum Cockpit Load Placard in Cockpit as well as entry in Flight Manual Page 9-1 have been checked/updated.

Equipment during weighing/calculation see Equipment List dated: _____

<Stamp> _____ Signature of Inspector

Erstellt: 25. FEB. 1988 <i>Yemba</i>	Geprüft: 25. FEB. 1988 <i>Uhepha</i>
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1.1 LOG OF REVISIONS

Revision No.	Pages affected	Description	LBA-approval signature	Date
1	1-1, 1-2, 4-2, 4-12, 4-13, 7-1	Hydraulic damper for aileron system (TB 6010/11, LBA-AD 86-140/2)		
2				
3	1-1, 1-2, 1-4, 2-4, 6-1 to 6-7	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)	 	8.02.1988

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2.4 MASS (WEIGHT)

Maximum take-off mass with water ballast 525 kg (1157 lbs)
without water ballast 380 kg (838 lbs)

Maximum landing mass 525 kg (1157 lbs)

Maximum mass of all non-lifting parts 230 to 245 kg (507-540 lbs)

Value must be determined according to table page 6-4, related to empty weight and empty weight C.G. position.

The term "non-lifting parts" includes the following: fuselage inclusive permanently fitted equipment, canopy and main pins plus maximum cockpit load. Tail fin water ballast is not counted for non-lifting parts, but for maximum weight.

Maximum wing water ballast mass 160 kg (352 lbs)
Loading instructions see page 9-2.

Maximum vertical tail fin water ballast mass 5.5 kg (12 lbs)
Loading instructions see page 9-3.

Maximum mass in Baggage Compartment 5 kg (11 lbs)
Loading instructions see page 4-6.

Maximum Cockpit Load 110 kg (242 lbs)

The term "Cockpit Load" includes the following: Pilot, parachute, baggage and temporary equipment.

Maximum cockpit load may be limited by mass of non-lifting parts. See entry on page 9-1.

Minimum Cockpit Load for club use (recommended)

Pilot and parachute 70 kg (154 lbs)
No baggage, no temporary equipment, no trim ballast

Pilot and parachute 55 kg (121 lbs)
3 trim weights fitted, no baggage, no temporary equipment

One trim weight (2.5 kg, 6 lbs) corresponds to 5 kg (11 lbs) of pilot mass.

If the sailplane does not fly in a club, it may be trimmed for higher minimum cockpit load. See instructions on page 9-4.

For minimum cockpit load see entry on page 9-1.

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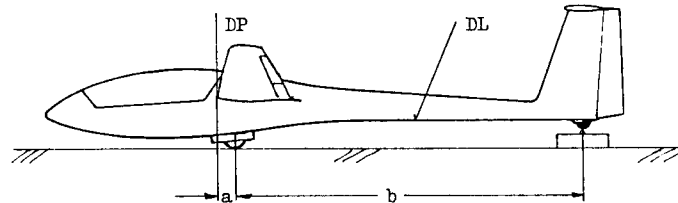
Erstellt: 25. FEB. 1988 <i>Heucke</i>	Geprüft: 25. FEB. 1988 <i>Whapke</i>
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6.1 WEIGHING PROCEDURE

To determine inflight C.G. position, the empty weight C.G. position must be known.

1. Determine total weight by weighing all parts and adding together. For inflight C.G. position, the pilot's weight must be added too.
2. Assemble sailplane. For inflight C.G., the pilot must be seated in the sailplane.
3. Raise tail on weighing machine until datum line is level using wooden blocks or adjustable jack. (Check with leveling gauge)
4. Determine gross tail weight.
5. Measure distance from tail support to center of landing gear axis.
6. Using plumb lead, determine points on floor perpendicular to left and right datum points, and points on floor perpendicular to center of landing gear axis. Measure distance <a> from wheel axis to datum point.



DATUM LINE <DL>: Under side of fuselage boom placed horizontal
 DATUM POINT <DP>: Leading edge of wing at root

7. Determine tare tail weight (Weight of auxiliary support used under 3)
8. Calculate nett tail weight = gross tail weight minus tare tail weight (the pilot was included):
9. Calculate empty weight C.G. position:

$$X_{cg} = \frac{\text{nett tail weight} * b}{\text{empty weight}} + a$$

10. Calculate loading limits according to page 6-2.

Form for Weighing Report see Instructions for Continued Airworthiness, page 8-4

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6.2 CALCULATION OF LOADING LIMITS

1. Determine Minimum Cockpit Load from Table "Empty Weight C.G. Position", page 6-6 or 6-7.
When being used in a club, Minimum Cockpit Load should be 70 kg (154 lbs). If it is higher, permanent ballast may be fitted under the forward seat portion, see page 9-1.

Finally resulting Minimum Cockpit Load should be entered in the following places:

- 1) in weighing report of inspection
 - 2) in Flight Manual page 9-1
 - 3) under instrument panel cover
 - 4) on Data Placard in cockpit
2. Maximum approved Weight of Non-lifting Parts may vary between 230 kg and 245 kg (507 lbs to 540 lbs).
In contrast to methods used up to now, Maximum Weight of Non-lifting Parts can be determined in relation to Empty Weight and Empty Weight C.G. Position according to table on page 6-4 and 6-5. See also example on page 6-3.

Maximum Weight of Non-lifting Parts should be entered into Weighing Report.

3. Determine Maximum approved Cockpit Load from table "Empty Weight C.G. Position", page 6-6 and 6-7.
Maximum Cockpit Load normally is 110 kg (242 lbs), as given in empty weight C.G. table. It may be lower due to trim conditions, excessive equipment or repairs.
Calculate Maximum Cockpit Load on weighing report, see also example on page 6-3.

Resulting Maximum Cockpit Load should be entered in the following places:

- 1) in weighing report of inspection
 - 2) in Flight Manual, page 9-1
 - 3) on Data Placard in cockpit
4. Empty weight (perhaps increased by weight of permanently fitted trim ballast) should be entered in the following places:
 - 1) in weighing report of inspection
 - 2) in Flight Manual, page 9-1 for calculation of maximum permissible water ballast weight

For permanent installation of trim ballast weights see page 9-5.

Form for Weighing Report see Instructions for Continued Airworthiness, page 8-4

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Heucke

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6.2 CALCULATION OF LOADING LIMITS continued

EXAMPLE for calculation of loading limits:

See also Instructions for Continued Airworthiness, form Weighing Report

1) Minimum Cockpit Load

For empty weight of 255 kg <562 lbs> and empty weight C.G. position 665 mm <26.18 in>, the Minimum Cockpit Load according to table page 6-6 and 6-7 is 75 kg <165 lbs> (Limit value ...667 mm <26.26 in> greater than actual value 665 mm <26.181 in>)

2) Maximum Weight of Non-lifting Parts

Maximum weight of non-lifting parts at empty weight of 255 kg <562 lbs> and empty weight C.G. position of 665 mm <26.181 in> is according to table page 6-4 and 6-5
..... 245 kg <540 lbs>

3) Maximum permissible Cockpit Load

FUSELAGE with complete equipment, battery, canopy and main pins	120.6 kg <266 lbs>
HORIZONTAL TAIL UNIT	6.3 kg < 14 lbs>
COCKPIT LOAD (Maximum 110 kg <242 lbs>	110 kg <242 lbs>
Weight of Non-lifting Parts	236.9 kg <522 lbs>
<hr/>	
MAXIMUM COCKPIT LOAD (Max. 110 kg <242 lbs>)	110 kg <242 lbs>

Maximum all-up weight 525 kg <1157 lbs>

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6.3 CALCULATION OF MAXIMUM WEIGHT OF NON-LIFTING PARTS

Maximum weight of non-lifting parts of 245 kg <540 lbs> must be reduced in relation to empty weight and empty weight C.G. position according to table below. (For lbs / inch values see following page)

Example: For empty weight C.G. position of 639 mm <25.16 in> and empty weight of 249 kg <549 lbs> the permissible weight of non-lifting parts is 232 kg <533 lbs>.

Empty Weight G <kg>	Empty weight C.G. Position Xs <mm>											
	from 540 to 559	from 560 to 579	from 580 to 599	from 600 to 619	from 620 to 639	from 640 to 659	from 660 to 679	from 680 to 699	from 700 to 719	from 720 to 739	from 740 to 759	from 760 to 779
246	230	230	230	231	232	233	234	236	237	238	239	240
247	230	230	230	231	232	233	235	236	237	238	239	241
248	230	230	230	231	232	233	235	236	237	238	240	241
249	230	230	230	231	232	234	235	236	237	238	240	241
250	230	230	230	231	232	234	235	236	237	238	240	241
251	230	230	230	231	233	234	235	236	237	238	240	241
252	230	230	230	231	233	234	235	236	238	239	240	241
253	230	230	230	231	233	234	235	236	238	239	240	241
254	230	230	230	232	233	234	235	237	238	239	240	241
255	230	230	230	232	233	234	235	237	238	239	240	242
256	230	230	231	232	233	234	235	237	238	239	240	242
257	230	230	231	232	233	234	236	237	238	239	241	242
258	230	230	231	232	233	234	236	237	238	239	241	242
259	230	230	231	232	233	235	236	237	238	240	241	242
260	230	230	231	232	233	235	236	237	238	240	241	242
261	230	230	231	232	233	235	236	237	239	240	241	242
262	230	230	231	232	234	235	236	237	239	240	241	242
263	230	230	231	232	234	235	236	238	239	240	241	243
264	230	230	231	232	234	235	236	238	239	240	241	243
265	230	230	231	233	234	235	236	238	239	240	242	243
266	230	230	231	233	234	235	237	238	239	240	242	243
267	230	230	231	233	234	235	237	238	239	241	242	243
268	230	230	232	233	234	235	237	238	239	241	242	243
269	230	230	232	233	234	236	237	238	239	241	242	243
270	230	230	232	233	234	236	237	238	240	241	242	244
271	230	230	232	233	234	236	237	238	240	241	242	244
272	230	231	232	233	235	236	237	238	240	241	242	244
273	230	231	232	233	235	236	237	239	240	241	243	244
274	230	231	232	233	235	236	237	239	240	241	243	244
275	230	231	232	233	235	236	237	239	240	241	243	244
276	230	231	232	234	235	236	238	239	240	242	243	244
277	230	231	232	234	235	236	238	239	240	242	243	244
278	230	231	232	234	235	236	238	239	241	242	243	245
279	230	231	232	234	235	237	238	239	241	242	243	245
280	230	231	233	234	235	237	238	239	241	242	243	245

6.3 CALCULATION OF MAXIMUM WEIGHT OF NON-LIFTING PARTS continued

Maximum weight of non-lifting parts of 540 lbs <245 kg> must be reduced in relation to empty weight and empty weight C.G. position according to table below. (For kg / mm values see preceding page)

Example: For empty weight C.G. position of 25.16 in <639 mm> and empty weight of 549 lbs <249 kg> the permissible weight of non-lifting parts is 511 lbs <234 kg>.

Empty Weight G<lbs>	Empty weight C.G. Position Xs <inches>											
	from 21.26 to 22.01	from 22.02 to 22.80	from 22.81 to 23.58	from 23.59 to 24.37	from 24.38 to 25.16	from 25.17 to 25.95	from 25.98 to 26.73	from 26.77 to 27.52	from 27.53 to 28.31	from 28.32 to 29.09	from 29.13 to 29.88	from 29.92 to 30.70
542	507	507	507	509	511	513	515	520	522	524	526	529
544	507	507	507	509	511	513	518	520	522	524	526	531
546	507	507	507	509	511	513	518	520	522	524	529	531
548	507	507	507	509	511	515	518	520	522	524	529	531
551	507	507	507	509	511	515	518	520	522	524	529	531
553	507	507	507	509	513	515	518	520	522	524	529	531
555	507	507	507	509	513	515	518	520	524	526	529	531
557	507	507	507	509	513	515	518	520	524	526	529	531
559	507	507	507	511	513	515	518	522	524	526	529	531
562	507	507	507	511	513	515	518	522	524	526	529	533
564	507	507	509	511	513	515	518	522	524	526	529	533
566	507	507	509	511	513	515	520	522	524	526	531	533
568	507	507	509	511	513	515	520	522	524	526	531	533
570	507	507	509	511	513	518	520	522	524	529	531	533
573	507	507	509	511	513	518	520	522	524	529	531	533
575	507	507	509	511	513	518	520	522	526	529	531	533
577	507	507	509	511	515	518	520	522	526	529	531	533
579	507	507	509	511	515	518	520	524	526	529	531	535
582	507	507	509	511	515	518	520	524	526	529	531	535
584	507	507	509	513	515	518	520	524	526	529	533	535
586	507	507	509	513	515	518	522	524	526	529	533	535
588	507	507	509	513	515	518	522	524	526	531	533	535
590	507	507	511	513	515	518	522	524	526	531	533	535
593	507	507	511	513	515	520	522	524	526	531	533	535
595	507	507	511	513	515	520	522	524	529	531	533	537
597	507	507	511	513	515	520	522	524	529	531	533	537
599	507	509	511	513	518	520	522	524	529	531	533	537
601	507	509	511	513	518	520	522	526	529	531	535	537
604	507	509	511	513	518	520	522	526	529	531	535	537
606	507	509	511	513	518	520	522	526	529	531	535	537
608	507	509	511	515	518	520	524	526	529	533	535	537
610	507	509	511	515	518	520	524	526	529	533	535	537
612	507	509	511	515	518	520	524	526	531	533	535	540
615	507	509	511	515	518	522	524	526	531	533	535	540
617	507	509	513	515	518	522	524	526	531	533	535	540

6.4 EMPTY WEIGHT C.G. TABLE <mm, kg>

For in, lbs values see following page

Empty Weight <kg>	Maximum Cockpit Load to Minimum Cockpit Load <kg>				
	110 - 70	110 - 75	110 - 80	110 - 85	110 - 90
246	659 - 659	659 - 677	659 - 695	659 - 713	659 - 730
247	657 - 658	657 - 676	657 - 694	657 - 711	657 - 729
248	656 - 657	656 - 675	656 - 693	656 - 710	656 - 728
249	654 - 656	654 - 674	654 - 691	654 - 709	654 - 726
250	653 - 655	653 - 673	653 - 690	653 - 708	653 - 725
251	651 - 654	651 - 672	651 - 689	651 - 706	651 - 724
252	649 - 653	649 - 671	649 - 688	649 - 705	649 - 723
253	648 - 652	648 - 669	648 - 687	648 - 704	648 - 721
254	646 - 651	646 - 668	646 - 686	646 - 703	646 - 720
255	645 - 650	645 - 667	645 - 685	645 - 702	645 - 719
256	643 - 649	643 - 666	643 - 683	643 - 700	643 - 717
257	642 - 648	642 - 665	642 - 682	642 - 699	642 - 716
258	640 - 647	640 - 664	640 - 681	640 - 698	640 - 715
259	639 - 646	639 - 663	639 - 680	639 - 697	639 - 714
260	637 - 645	637 - 662	637 - 679	637 - 696	637 - 713
261	636 - 644	636 - 661	636 - 678	636 - 695	636 - 711
262	634 - 643	634 - 660	634 - 677	634 - 694	634 - 710
263	633 - 642	633 - 659	633 - 676	633 - 692	633 - 709
264	631 - 642	631 - 658	631 - 675	631 - 691	631 - 708
265	630 - 641	630 - 657	630 - 674	630 - 690	630 - 707
266	628 - 640	628 - 656	628 - 673	628 - 689	628 - 706
267	627 - 639	627 - 655	627 - 672	627 - 688	627 - 704
268	625 - 638	625 - 654	625 - 671	625 - 687	625 - 703
269	624 - 637	624 - 653	624 - 670	624 - 686	624 - 702
270	623 - 636	623 - 653	623 - 669	623 - 685	623 - 701
271	621 - 635	621 - 652	621 - 668	621 - 684	621 - 700
272	620 - 634	620 - 651	620 - 667	620 - 682	620 - 699
273	619 - 634	619 - 650	619 - 666	619 - 682	619 - 698
274	617 - 633	617 - 649	617 - 665	617 - 681	617 - 697
275	616 - 632	616 - 648	616 - 664	616 - 680	616 - 696
276	614 - 631	614 - 647	614 - 663	614 - 679	614 - 694
277	613 - 630	613 - 646	613 - 662	613 - 678	613 - 693
278	611 - 629	611 - 645	611 - 661	611 - 677	611 - 692
279	610 - 629	610 - 644	610 - 660	610 - 676	610 - 691
280	608 - 628	608 - 643	608 - 659	608 - 675	608 - 690

6.4 EMPTY WEIGHT C.G. TABLE <in, lbs>

For mm, kg values see preceding page

Empty Weight <lbs>	Maximum Cockpit Load to Minimum Cockpit Load <lbs>				
	232 - 154	232 - 165	232 - 176	232 - 187	232 - 198
542	25.945- 25.945	25.945- 26.654	25.945- 27.362	25.945- 28.071	25.945- 28.740
544	25.866- 25.906	25.866- 26.614	25.866- 27.323	25.866- 27.992	25.866- 28.701
546	25.827- 25.866	25.827- 26.575	25.827- 27.283	25.827- 27.953	25.827- 28.661
548	25.748- 25.827	25.748- 26.535	25.748- 27.205	25.748- 27.913	25.748- 28.583
551	25.709- 25.787	25.709- 26.496	25.709- 27.165	25.709- 27.874	25.709- 28.543
553	25.630- 25.748	25.630- 25.400	25.630- 27.126	25.630- 27.795	25.630- 28.504
555	25.551- 25.709	25.551- 26.417	25.551- 27.087	25.551- 27.756	25.551- 28.465
557	25.512- 25.669	25.512- 26.339	25.512- 27.047	25.512- 27.717	25.512- 28.386
559	25.433- 25.630	25.433- 26.299	25.433- 27.008	25.433- 27.677	25.433- 28.346
562	25.394- 25.591	25.394- 26.260	25.394- 26.969	25.394- 27.638	25.394- 28.307
564	25.315- 25.551	25.315- 26.220	25.315- 26.890	25.315- 27.559	25.315- 28.228
566	25.276- 25.512	25.276- 26.181	25.276- 26.850	25.276- 27.520	25.276- 28.189
568	25.197- 25.472	25.197- 26.142	25.197- 26.811	25.197- 27.480	25.197- 28.150
570	25.157- 25.433	25.157- 26.102	25.157- 26.772	25.157- 27.441	25.157- 28.110
573	25.079- 25.394	25.079- 26.063	25.079- 26.732	25.079- 27.402	25.079- 28.071
575	25.039- 25.354	25.039- 26.023	25.039- 26.693	25.039- 27.362	25.039- 27.992
577	24.961- 25.315	24.961- 25.984	24.961- 26.654	24.961- 27.323	24.961- 27.953
579	24.921- 25.276	24.921- 25.945	24.921- 26.614	24.921- 27.284	24.921- 27.913
582	24.843- 25.276	24.843- 25.906	24.843- 26.575	24.843- 27.245	24.843- 27.874
584	24.803- 25.236	24.803- 25.866	24.803- 26.535	24.803- 27.165	24.803- 27.835
586	24.724- 25.197	24.724- 25.827	24.724- 26.496	24.724- 27.126	24.724- 27.795
588	24.685- 25.157	24.685- 25.787	24.685- 26.456	24.685- 27.087	24.685- 27.756
590	24.606- 25.118	24.606- 25.748	24.606- 26.417	24.606- 27.047	24.606- 27.717
593	24.567- 25.079	24.567- 25.709	24.567- 26.378	24.567- 27.008	24.567- 27.677
595	24.528- 25.039	24.528- 25.709	24.528- 26.339	24.528- 26.969	24.528- 27.638
597	24.449- 25.000	24.449- 25.669	24.449- 26.299	24.449- 26.929	24.449- 27.599
599	24.409- 24.961	24.409- 25.630	24.409- 26.260	24.409- 26.890	24.409- 27.560
601	24.370- 24.961	24.370- 25.591	24.370- 26.220	24.370- 26.850	24.370- 27.520
604	24.291- 24.921	24.291- 25.551	24.291- 26.181	24.291- 26.811	24.291- 27.480
606	24.252- 24.882	24.252- 25.512	24.252- 26.142	24.252- 26.772	24.252- 27.441
608	24.173- 24.843	24.173- 25.472	24.173- 26.102	24.173- 26.732	24.173- 27.401
610	24.134- 24.803	24.134- 25.433	24.134- 26.063	24.134- 26.693	24.134- 27.361
612	24.055- 24.764	24.055- 25.394	24.055- 26.023	24.055- 26.654	24.055- 27.321
615	24.016- 24.764	24.016- 25.354	24.016- 25.984	24.016- 26.614	24.016- 27.281
617	23.937- 24.723	23.937- 25.315	23.937- 25.945	23.937- 26.575	23.937- 27.241

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1.1 LOG OF REVISIONS

Revision No.	Pages affected	Description	LBA-approval signature	Date
1	1-1, 1-2, 8-4	Method to determine permissible weight of non-lifting parts modified analogous to LS6-b (TB 6015)		

LS6-a Manuals can be ordered from:
 ROLLADEN-SCHNEIDER Flugzeugbau GmbH
 Mühlstrasse 10
 D-6073 Egelsbach
 Federal Republic of West Germany

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Erstellt: 25. FEB. 1988 <i>Leu ke</i>	Geprüft: 25. FEB. 1988 <i>hapha</i>
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(6AWE1)

Serial No. _____ Reg.Signs _____ Date: _____ **WEIGHING REPORT**

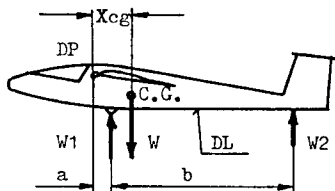
COMPONENT WEIGHTS (Check when equipment altered or every fourth year)

	Dimension		Dimension
Right Wing		Maximum all-up Weight (1157 lbs)	525 kg
Left Wing		Maximum Weight of Non-lifting Parts at W= . . . kg/lbs and Xcg= . . . mm/in according to table on page 6-4/6-5 of Flight Manual.	
Fuselage + Equip- ment + Main Pins + Canopy		Fuselage (as on left side)	
Horizontal Tail		Horizontal tail	
Empty Weight <W>		Cockpit Load (max.110 kg or 242 lbs)	
BATTERY None		Weight of Non-lifting Parts	
POSITION Baggage Compartment		FIXED BALLAST POS. . . . kg/lbs Tail Fin/. . .	
Tail Fin			

WEIGHING AND EMPTY WEIGHT C.G. DETERMINATION (Check when equipment altered or every fourth year)

Technical data according to TCDS: Datum Point <DP>: Leading edge of wing at root
Datum Line <DL>: Under side of fuselage boom horizontal

Empty Weight <W>		Distance Wheel axis - Datum Point <a>	
Nett Tail Weight <W2>		Distance Wheel axis - Tail support 	



$$\frac{W2 * b}{W} + a = Xcg$$

* + = kg/lbs

NOTE: See also Flight Manual Chapter 6

State dimensions used. Redetermine distances a and b, because of possibly altered suspension level.

Empty Weight C.G. Range according to Flight Manual, Chapter 6 from . . . mm/in to . . . mm/in at

Empty Weight <W> . . . kg/lbs yields permissible Cockpit Load Range

from . . . kg/lbs to . . . kg/lbs. Weighed and calculated C.G. position is within permissible limits.

MINIMUM COCKPIT LOAD: _____ kg/lbs **MAXIMUM COCKPIT LOAD:** _____ kg/lbs

Weight and Balance Placard and Minimum Cockpit Load Placard in Cockpit as well as entry in Flight Manual Page 9-1 have been checked/updated.

Equipment during weighing/calculation see Equipment List dated:

<Stamp> _____ Signature of Inspector

EDITION: 25.Feb.88

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Erstellt: 25. FEB. 1988 <i>Heuck</i>	Geprüft: 25. FEB. 1988 <i>Whapha</i>
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